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AN ANALYSIS OF THE SKIPPING CHARACTERISTICS
OF SOME FULL-SIZE FLYING BOATS

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Navy Department

NACA

WASHINGTON

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ADVANCE RESTRICTED REPORT NO. 5J24

AN ANALYSIS OF THE SKIPPING CHARACTERISTICS
OF SOME FULL-SIZE FLYING BOATS

By F. W. S. Locke, Jr.

SUMMARY

An analysis is made of the skipping characteristics of some full-size flying boats. In many cases the only source of data was pilot opinion, and sometimes this was contradictory. It was found that the more experience a pilot had with a particular airplane, the more apt he was to be untroubled by the skipping characteristics of that airplane.

Since it is currently desirable to design for inexperienced pilots, a figure of merit was adopted which is in inverse proportion to the amount of experience required to make stable landings. Using this basis for analysis, it has been found that the skipping characteristics can be improved by

1. Decreasing the initial load coefficient
2. Increasing the step height
3. Decreasing the sternpost angle

The load coefficient and the sternpost angle will ordinarily be chosen from other considerations. Figure 4 may then be used to determine a step height which should insure freedom from good skipping.

There are other unresolved hull form parameters which are known from model tests to have large secondary effects on skipping. Hence, use of the dashed line in figure 4 may be considered as an upper limit for purposes of good design.

INTRODUCTION

Skipping may be defined as an unstable oscillation of hydrodynamic

origin, predominantly heaving in character, which can occur just after landing or just prior to take-off. Skipping is associated with conditions whereby the forebody carries most of the water-borne load and at the same time a large amount of water is washing over the afterbody bottom. A good description of the mechanism of skipping may be found in reference 1.

Too numerous a series of serious accidents to both large and small flying boats has been traced to skipping. Because of the fact that hulls with quite violent skipping tendencies can be safely handled by experienced and careful pilots, there has been, in the past, some tendency on the part of designers to minimize the inherent danger. The accumulated experience of the average pilot is decreasing during the war; furthermore, there is tactical need for operations under increasingly adverse conditions. New airplanes should be designed, therefore, to take into account these two factors which have an important effect on the landing characteristics of a long-range military flying boat. Not only may the pilot be relatively inexperienced, but he also may be fatigued or wounded. Further, because of lack of fuel at the end of a long flight, it may be necessary to set down in rough water. Hence, it would seem to be very important for new military flying boats to have especially good landing characteristics under adverse conditions.

The purpose of this report is to gather together in one place information on the skipping characteristics of a number of flying boats. A graph showing the influence of the hull form on the skipping characteristics is given which should be useful in laying out the proportions of the hull in preliminary design to insure good landing behavior. Upper-limit porpoising is not considered in this report.

DATA

In tables I and II, the pertinent particulars and specifications, and a single word describing the skipping characteristics are given for each hull. In some cases, this one word represents an interpretation of a flight-test report, and in others the interpretation of opinions expressed by different pilots who had flown the particular airplane. In a few cases, diametrically opposed views were obtained from different pilots. The viewpoint appears to be in function of the amount of experience the dissenting pilot has had with the airplane. A good example is the Catalina, about which one pilot said: "It can be landed under any circumstance without skipping." He had had a great deal of experience with them. Another pilot, who had a great deal of experience with Mariners, said: "I have the 'willies' whenever I have to land a Catalina." Actually, neither pilot is entirely wrong. The first man was able to make stable landings instinctively by virtue of experience

which the latter pilot did not have. A third pilot, who liked the Catalina, stated he could land it without skipping any time he wished, but that it would skip very unpleasantly when landed improperly. This man had more experience with different types of flying boats than either of the other two. It seemed proper, then, to rate the skipping characteristics of this flying boat as only "fair."

Similarly, the other hulls are rated as "good" when, under most circumstances, they could make stable landings. The "poor" hulls are those which, in most cases, are unstable on landing. In nearly all cases, both stable and unstable landings are possible depending on pilot technique. The method of rating, therefore, is to some degree the inverse of the amount of pilot experience required to make stable landings.

The behavior on landing was chosen as the criterion of comparison because, on the basis of experience, it seems to be somewhat easier to induce skipping accidentally on landing than on take-off. Any flying boat that showed any evidence of inadvertent skipping on take-off, therefore, was automatically rated as poor.

The data are not really quantitative. A method of making the results quantitative would be to determine the range of stable and unstable landing trims for all the hulls listed. This should enable a much better correlation of skipping with hull form. An appendix gives such detailed information as could be found about each flying boat. It is not believed to be sufficiently complete to improve materially the interpretation of the data.

Throughout this report the following notation and nondimensional coefficients are used:

C_{Δ_0} initial load coefficient
 L_F/b forebody length coefficient
 L_R/b afterbody length coefficient
 h step height
 σ sternpost angle

where

Δ_0 initial load on the water, pounds
 w specific weight of water, pounds per cubic foot (64.0 for sea water)

- b beam at main step, feet
- L_f forebody length, measured from the intersection of chine and keel to the step centroid along a line parallel to the tangent to the forebody keel, feet
- L_a afterbody length, measured from the step centroid to the second step or sternpost along a line parallel to the tangent to the forebody keel, feet
- h step height, measured at the step centroid, percent of beam at step
- σ sternpost angle, the angle between the tangent to the forebody keel at the main step and a line joining the tip of the step and the sternpost, degrees

Figure 2 defines the principal dimensions used. Considerable effort has been made to bring the dimensions, given on tables I and II, into conformation with these definitions. Unfortunately, errors may still exist because it was found on several occasions the drawings of the same airplane issued by the same manufacturer would be inconsistent. However, it is believed that the dimensions given here are a good deal more accurate than those given in reference 2 and at least as accurate as, if not more so than, those given in reference 3.

ANALYSIS

At the outset, it appeared that the get-away speed coefficient might be the primary independent variable (reference 1). However, it does not make a very satisfactory variable for correlating full-scale behavior because it depends on extraneous things like flap setting, wind, and pilot technique. Wing loading would eliminate these things, but it has the important disadvantage that it is not clearly related to the hull. On the other hand, the initial load coefficient C_{Δ_0} is a more suitable variable since empirically it is a function of the get-away speed. (See reference 2.) It varies when the hull size is varied with a given wing and gross weight, or when the gross weight changes in a given airplane. Hence, C_{Δ_0} has been used as the primary independent variable. For studies of systematic model experiments it would not be nearly so satisfactory as it is here.

The first step taken was to plot the step height h against the initial load coefficient C_{Δ_0} . The result is shown in figure 3, where

the different skipping behaviors are differentiated by different symbols. A line which has been cross-hatched on the undesirable side was drawn to put a few "good" points below and as few "fair" or "poor" points above the line as possible. It is evident that this line is not very effective in "separating the sheep from the goats." The next step was to plot σ against C_{Δ_0} in various ways. Figure 4 shows that by using h/σ some improvement is achieved over the correlation shown in figure 3.

Further plots, which introduced afterbody length and get-away speed coefficient in various fashions as additional parameters, were tried; these indicated no systematic effect of either variable. Since a fairly good correlation had already been obtained in figure 4, the matter was not pursued further.

DISCUSSION

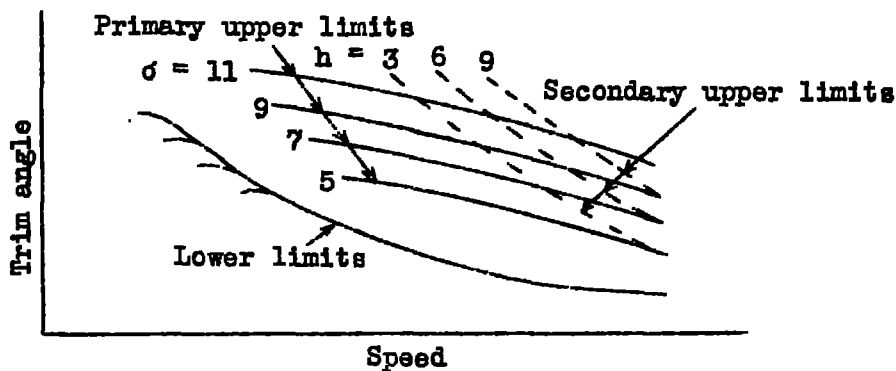
The limitations of the data should be kept firmly in mind. It is more or less quantitative, being, as far as possible, the average of more than one opinion. It cannot, however, be very precise by its very nature. Even with the hulls classed as "poor," stable landings in the hands of a capable and experienced pilot are possible. Thus, it is difficult to lay down a hard-and-fast line, differentiating between the hulls having "good" and "poor" skipping characteristics. This is especially true since there are known to be secondary variables of considerable importance. It is believed, however, that the dashed line in figure 4 should provide a safe basis for preliminary design.

Figure 3 merely reaffirms all previous experience (references 1, 3, 4, and 5) that the step height has a powerful influence on the skipping characteristics. It seems clear, however, that there has not been sufficient delineation of the "poor" from the "good" hulls to allow using this chart for design purposes. A dashed line was drawn in parallel to the cross-hatched line so that the XPBB-1 point lay above the line, because this flying boat has outstandingly good skipping characteristics. The line serves to emphasize the necessity of deep steps.

Figure 4 shows a quite good correlation of skipping characteristics with hull form. It indicates that, if the sternpost angle is fixed from other considerations, then the step height for good skipping characteristics may be found, and that the higher the sternpost angle the higher the step height must be to give desirable skipping characteristics. At first glance, this is surprising in view of Parkinson's "jet" conception (reference 1) from which it might be expected that, with a fixed step height, skipping would be reduced by increasing the sternpost angle in-

stead of harming it. However, model tests (reference 6) actually indicate that the skipping characteristics are somewhat harmed by increasing the sternpost angle. Since Parkinson's theory gives such a reasonable explanation of step height and ventilation, it is thought that the influence of the sternpost angle must be largely confined to the rear half of the afterbody.

The explanation may lie in the following schematic sketch, which is intended to show the independent effect of both sternpost angle and step height on the stability limits.



Model data in reference 5 indicates that the primary upper limit is unaltered by changes of step height alone. On the other hand, model data in reference 6 shows that the variations of the sternpost angle cause only relatively small changes in the position of the secondary upper limit. There is apparently a region of speeds and trims in which skipping is possible with a given step height and moderately large afterbody angle. As the sternpost angle is reduced, the primary upper limit covers up more and more of this region, until a point is reached below which no further improvement in the skipping characteristics should be expected. Hence, the preceding sketch may be taken as at least an indication of the combined influences of step height and sternpost angle, and is an extension of similar sketch in reference 7. It does not seem unreasonable to push the ideas expressed in reference 7 a little further and state that the rear half of the afterbody is largely responsible for the primary upper limit, and the forward half for the secondary upper limit, and skipping characteristics. This conception is probably a valid explanation of figure 4. It is believed that this brings about only a clarification of the ideas expressed in references 1 and 3, and that Parkinson's "jet" theory offers a broad explanation of the mechanism of skipping.

The sternpost angle ordinarily will be chosen to give as high a primary upper limit of stability as is compatible with hump trim angles and main spray characteristics. The step height which should ensure

good skipping characteristics can then be found with the aid of the recommended line in figure-4. The step height found in this manner will be a great deal deeper than has been ordinarily used in past practice. It will, unfortunately, cause a considerable amount of air drag. By keeping in mind Parkinson's jet theory, which requires that air be admitted to relieve the low pressures created by the high-speed jet issuing from the step, there appear to be several ways of reducing the step height while retaining good skipping characteristics.

The first method is by using a step of moderate depth and ventilation ducts running into the afterbody bottom. The area of the required ducts is quite large (reference 1) and they may interfere seriously with the interior arrangements of the flying boat. Ventilation ducts have been used on several aircraft which already had poor skipping characteristics; when sufficient duct area was used, they seemed to have a beneficial effect on the skipping characteristics.

A better method of reducing the necessary step height is to use a V-plan-form step (reference 8). As may be seen from table I, the V-plan-form step has been quite widely used in the past. If an ordinary transverse step is transformed into a V-step, as on the PBN-1, the change may accomplish very little. On the other hand, if the hull is designed for a V-step, as on the Short "Empire," and if it is not too shallow, the V-step can be superior to the ordinary transverse step. To illustrate the differences in the V-steps of these two airplanes, the sketches in figure 1 are worth noting. On the "Empire" boat, the face of the V-step is of uniform depth, while on the PBN-1 there is a definite throttle near the chine. Model and full-scale experiments on ventilation (references 1 and 9) have shown that the introduction of air at the keel is needed to alleviate skipping. From an examination of the PBN-1, because of the throttle near the chine, it seems obvious that air would have a great deal of difficulty reaching the keel.

Since the V-step is believed to be fundamentally superior to the transverse step, an opportunity arises for aerodynamic fairing. British tests (references 3 and 10) on the Sunderland III have indicated that by using a moderate fairing the water stability is not appreciably altered. It is also reported that such a fairing increases the cruising speed about 5 miles per hour, which is an exceedingly large benefit.

Afterbody length, step plan form, and the difference between the forebody and afterbody dead rise near the step undoubtedly influence the skipping characteristics. It is not easy to disentangle their separate influences from the available full-scale data. Reference 3 should be consulted for a summary of model and full-scale evidence on these and other points. It would appear that the only satisfactory method of determining the individual effect of other hull-form variables is through systematic model tests.

One of the things which keeps recurring in the British reports concerning skipping (references 9, 11, and 12, for instance) is that disturbed water aggravated the skipping very materially. The Sea Otter (reference 9) is apparently stable during take-off, but violent skipping was set up when it ran across the wake of a boat. Reference 4 recounts some of the model experience at Stevens Institute of Technology on the influence of rough water on skipping. In another case a model that seemed quite stable could be induced into a violent self-sustaining skipping by allowing it to run through three of four waves. These experiences serve to emphasize the necessity of exceptionally good skipping characteristics, so that rough-water landings can be performed safely.

CONCLUSIONS

On the basis of available full-scale evidence bearing on the skipping characteristics of flying boats, step height, sternpost angle and load coefficient appear to be the major variables.

The sternpost angle should be selected so as to obtain the best compromise involving the primary upper limit of stability, hump trim angles, and the main spray characteristics. When the sternpost angle has been selected, the step height may be found by using the dashed line on figure 4.

The shape and location of the rear half of the afterbody control the location of primary upper limit. The shape and location of the forward half of the afterbody control the location of the secondary upper limit and the skipping characteristics. These two bald statements are believed to be in substantial agreement with the conceptions expressed in references 1 and 3.

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APPENDIX A

DETAILS OF THE SKIPPING CHARACTERISTICS OF VARIOUS FLYING BOATS

This appendix gives more information on the skipping characteristics of the various flying boats listed in tables I and II. The purpose of this greater detail is primarily to justify the "merit" in which each hull was classed. It also should be useful in that it gathers together in one place available information on the various hulls. More information on the dimensions of some of the airplanes may be found with the aid of reference 2.

Amphibians P-III-BPoor

The following is quoted from a hydrodynamic data report by C. E. Kahlka, Jr., in 1942: "It was usually possible to check oscillations during the high-speed portion of the run before they acquired appreciable amplitude, but at times this was decidedly tricky and involved skipping over the water surface much like a skipping stone."

Grumman J4F-2Good

As far as could be determined, no trouble has been experienced with skipping. The airplane apparently has a tendency to nose over when landed at too low a trim angle, but this is more likely to be connected with the low dead-rise bow sections and convex curved water planes of the forebody, than with the skipping characteristics.

Vought OS2U-1Good

This airplane has a considerable amount of lower-limit porpoising during take-off unless skillfully handled. No reports of skipping on landing have been found, however.

Saro 37GoodReferences 3 and 11

Fairly complete tests indicated some danger of skipping if the pilot allowed the airplane to get into abnormal attitudes. The worst condition appeared to be the full-stall landing; otherwise landings were stable. Rough water aggravates matters.

Fairey SealPoorReference 3

This twin-float seaplane apparently has no stable range or else a very narrow one. Conditions are worse in rough water than in smooth.

Grumman J2F-5Fair

According to pilot's opinions, full-stall and flat-power-on landings are stable, but there is a rather wide range of intermediate angles which are unstable.

Curtiss XSC-1Good

Preliminary reports indicate good landing characteristics.

Grumman JRF-4FairReference 1

Full-stall landings are quite stable, and intermediate angles moderately unstable. Fast, flat, power-on landings are very unstable. The landing stability of this aircraft has been considerably improved by moving the step aft, thereby increasing the step height. Landings over quite a wide range of trims are now stable. The altered aircraft is called the JRF-5.

Supermarine Sea OtterPoorReference 9

A little rough water would set up violent skipping during take-off. Structural damage to the engine mount was experienced after skipping during one landing.

Hall PH-3Good

Full-stall and moderate angle landings quite stable according to one pilot. This may well be partially attributable to the very low wing loading.

Consolidated PBV-5Fair

Full-stall and flat power-on landings are stable, but there is a rather wide range intermediate angle which is moderately unstable.

Naval Aircraft Factory FBN-1Fair

Landings made at low trims are quite stable, but some skipping encountered at higher trims. Full stall is unstable. Ventilation, using 1.7 square feet of ventilation area, helped a little. There is disagreement as to whether this airplane is better or worse than the PBX, from which it was derived.

Short Singapore IIc and IIIGoodReference 3

Behavior quite similar to the Hall PH-3, of which the Singapore is a contemporary. The stable range of trims for this hull is about 8° wide.

Saro LerwickReference 13

According to reference 13, the shallow transverse step was exceeding unstable full-scale; after lowering the forebody and incorporating a V-step, a full-scale stable form was produced. The modified hull has almost 11 percent step depth at the keel. The results shown in reference 3 are believed to apply to the original form.

Short Sunderland IGoodReference 14

The airplane has a wide range of attitudes for stable landings. From about 4° to 9° trim angles are stable.

Consolidated 31Poor

Consolidated flight tests indicated skipping during both take-off and landing.

Consolidated XP4Y-1Fair

Consolidated flight tests indicated a very marked improvement, though apparently there was still some skipping on landing.

Martin PBM-3Good

It is possible to land this aircraft over a wide range of trims without any difficulty from skipping.

Short Sunderland IIIGoodReference 10

Even with a 1:4 step fairing, deliberate attempts to get skipping were quite unsuccessful. In reference 10, it is stated that by making violent full-stall landings an occasional skip would be encountered, but not enough to define a stability limit. Service usage has indicated that the full-stall landing is unstable in slightly rough water. However, a fairly large number of the aircraft have been in service for about four years, and by landing at intermediate angles have had no trouble from skipping.

Martin PBM-5Fair

The aircraft has been considerably overloaded. Mild skipping can be found between trim angles of 10° and 14° . The intensity of this skipping increases with increasing gross weight.

Boeing XPBB-1GoodReference 1

It is possible to get upper-limit porpoising with this flying boat, but, as far as is known, there is no skipping on landing at any attitude or load. It is looked upon with great enthusiasm, as far as skipping is concerned, by all pilots who have flown it. It is believed to be the most stable aircraft in service today.

Consolidated PB2Y-3PoorReference 1

Violent skipping was found over a wide range of trims. After production had started, the skipping was alleviated by the introduction of ventilation ducts. Some pilots have stated, under controlled test conditions, that they could not tell from the instability on landing whether the ducts were open or closed. Other pilots have reported considerable improvement due to the step ventilation.

Short GFairReferences 3 and 16

At low attitudes no instability occurs on landing, but, as the trim increases, skipping is progressively introduced until, at high attitudes, it is quite bad. According to reference 16 no trouble need be had if the airplane is landed fast with a little power.

Boeing 314Reference 1

The flying boat was violently unstable just after landing at high trims. By moving the step aft, thereby increasing the depth because of the angle between the forebody and afterbody keels, a great improvement in landing at high trims resulted.

Martin XPB2M-1Fair

At trim angles below about 6° this airplane is stable. Above this trim, quite violent skipping can be obtained. Because of the large size of the airplane, the motion is quite slow and is very easy to control and, hence, may not be objectionable to the pilot. However, it should be noted that a large number of the pilots have been men with years of flying-boat experience. Model experience with this airplane (reference 3) indicates that it is not so good as it could be with a deeper step.

REFERENCES

1. Parkinson, John B.: Notes on the Skipping of Seaplanes. NACA RB No. 3127, 1943.
2. Locke, Fred W. S., Jr.: A Correlation of the Dimensions, Proportions, and Loadings of Existing Seaplane Floats and Flying-Boat Hulls. NACA ARR, March 1943.
3. Smith, A. G., and White, H. G.: A Review of Porpoising Instability of Seaplanes. Rep. No. H/Res/173, NAAE, Feb. 29, 1944.
4. Truscott, Starr, and Olsen, Roland E.: The Longitudinal Stability of Flying Boats as Determined by Tests of Models in the NACA Tank. II - Effect of Variations in Form of Hull on Longitudinal Stability. NACA ARR, Nov. 1942.
5. Locke, F. W. S., Jr., and Hugli, W. C., Jr.: A Method for Studying the Longitudinal Dynamic Stability of Flying-Boat-Hull Models at High Planing Speeds and during Landing. NACA ARR No. 4H31, 1944.
6. Land, Norman S., and Linn, Lindsay J.: Tests of a Dynamic Model in NACA Tank No. 1 to Determine the Effect of Length of Afterbody, Angle of Afterbody Keel, Gross Load, and a Pointed Step on Landing and Planing Stability. NACA ARR, March 1943.

7. Davidson, Kenneth S. M., and Locke, F. W. S., Jr.: Some Analyses of Systematic Experiments on the Resistance and Porpoising Characteristics of Flying-Boat Hulls. NACA ARR No. 3I06, 1943.
8. Parkinson, John B., and Land, Norman S.: The Landing Stability of a Powered Dynamic Model of a Flying Boat with a 30°V-Step and with Two Depths of Transverse Step. NACA RB No. 4B14, 1944.
9. Fletcher, G. L.: Tank Tests on the Water Stability of the Supermarine Sea Otter. Rep. No. Aero 1927, R.A.E., March 1944.
10. Naylor, P. E.: The Effect of Weight Variation on the Water Stability, Trim and Elevator Effectiveness of a Sunderland III Aircraft during Take-Off and Landing. Rep. No. H/Res/171, MAEE, Feb. 19, 1944.
11. White, H. G., and Smith, A. G.: Stability Tests on a Large Scale Model of the Shetland Hull Bottom. Rep. No. H/Res/159, MAEE, Dec. 1942.
12. Gott, J. P.: Note on the Technique of Tank Testing Dynamic Models of Flying Boats as Affected by Recent Full Scale Experience. Rep. B. A. 1572, RAE, Dec. 1939.
13. MacPhail, D. C., Tye, W. D., and Llewelyn-Davies, D. I. T. P.: Model Investigation on the Effect of Slipstream on the Longitudinal Stability of Flying Boats on the Water. Rep. No. Aero 1792, RAE, Jan. 1943.
14. White, H. G., Smith, A. G., and Shaw, R. A.: Attitude and Stability Measurements on a Half Scale Sunderland Hull Fitted to the Scion Seaplane. Rep. No. H/Res/158, MAEE, Dec. 1942.
15. Shaw, R. A., and Evans, G. J.: Porpoising Tests on the "G" Class Flying Boat. Rep. No. H/Res/141, MAEE, March 1941.

TABLE I.- SKIPPING CHARACTERISTICS OF VARIOUS FLYING BOATS

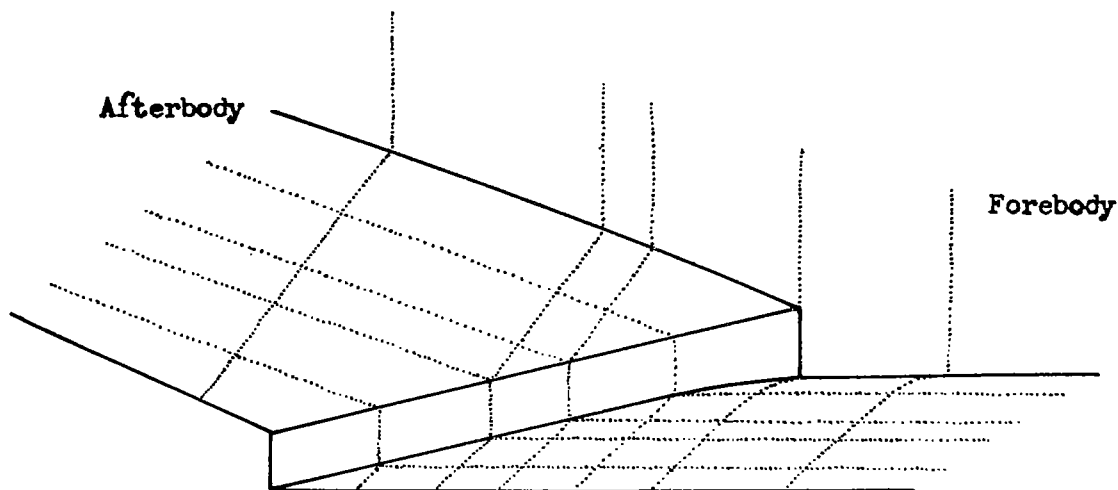
Manufacturer	Model	Gross weight (lb)	Wing area (ft ²)	Beam (ft)	C _{Δo}	Step		$\frac{L_f}{b}$	$\frac{L_b}{b}$
						Height percent b	Shape ¹		
Amphibians	P-IIIB	3,200	224	4.66	0.493	4.5	T	2.05	1.92
Grumman	J4F-2	4,500	245	4.25	.915	6.6	T	3.13	2.12
Vought	OS2U-1	4,841	261	3.67	1.530	7.8	T	3.85	3.08
Saro	37	5,700	340	4.25	1.160	10.0	20°FV	3.60	3.73
Fairey	Seal	6,400	445	3.20	1.525	6.5	T	4.22	3.37
Grumman	J2F-5	6,651	409	5.00	.830	5.0	T	2.86	2.69
Curtiss	XSC-1	7,642	281	4.38	1.420	8.8	T	3.64	3.04
Grumman	JRF-4	8,000	375	5.00	1.000	5.9	T	2.84	2.33
Grumman	JRF-5	8,000	375	5.00	1.000	7.3	T	2.93	2.24
Supermarine	Sea Otter	10,200	610	5.17	1.153	4.0	T	3.20	3.90
Hall	PH-3	16,150	1170	8.35	.435	3.35	T	2.90	1.69
Consolidated	PBY-5	26,000	1400	10.21	.380	3.5	T	2.44	1.51
Short	Singapore II	26,600	1760	10.70	.339	4.6	30°V	2.32	2.16
NAF	PBM-1	28,000	1400	10.19	.415	3.5	20°V	2.70	1.94
Short	Singapore III	31,150	1760	10.70	.397	4.6	30°V	2.32	2.16
Saro	Lerwick Ia	33,000	845	8.48	.845	3.4	T	3.04	3.69
Saro	Lerwick Ib	33,000	845	8.48	.845	7.2	30°V	3.16	3.67
Short	Sunderland I	43,000	1487	9.10	.892	9.7	30°V	3.47	3.36
Consolidated	31	46,000	1048	9.17	.930	2.4	T	3.01	2.51
Consolidated	XP4Y-1	46,000	1048	9.17	.930	4.9	30°V	2.94	2.58
Martin	PBM-3	46,500	1406	10.00	.725	5.0	T	3.28	2.74
Short	Sunderland III	50,000	1487	9.10	1.035	9.7	30°FV	3.47	3.36
Martin	PBM-5	56,000	1406	10.00	.873	5.0	T	3.28	2.74
Boeing	XPBB-1	62,000	1826	10.42	.855	9.0	T	3.40	2.67
Consolidated	PB2Y-3	68,500	1780	10.50	.925	3.5	30°V	2.85	2.14
Short	G	73,500	2390	11.65	.726	5.7	30°V	3.10	3.45
Boeing	314	82,500	2867	12.50	.660	3.0	T	3.60	2.12
Boeing	A-314	82,500	2867	12.50	.660	4.75	T	3.75	1.97
Martin	XPB2M-1R	140,000	3683	13.50	.890	5.0	T	3.32	2.75

¹ T - transverse step; V - V-step; FV - faired V-step.

TABLE II.- SKIPPING CHARACTERISTICS OF VARIOUS FLYING BOATS

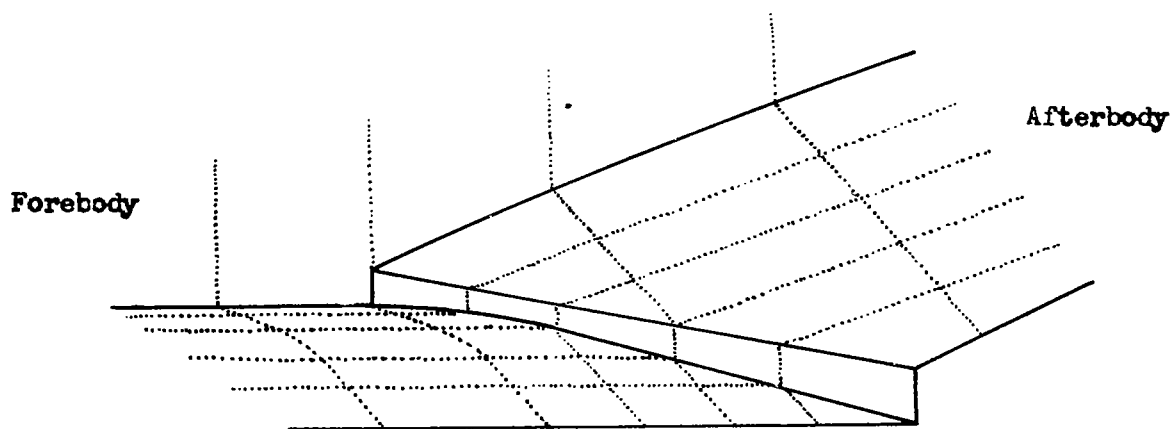
Model	Dead rise ¹		Sternpost angle (deg)	Landing merit	Source
	Forebody (deg)	Afterbody (deg)			
P-IIIB	7	7	12.0	Poor	Owner's flight tests
J4F-2	20	20	7.0	Good	Opinions
OS2U-1	25	25	7.8	Good	Opinions
37	27	37	9.0	Good	RAF flight tests
Seal	37	37	7.6	Poor	RAF flight tests
J2F-5	25	25	9.0	Fair	Opinions
XSC-1	Fluted	Fluted	8.6	Good	Opinions
JRF-4	25	25	9.1	Fair	NACA RB No. 3I27
JRF-5	25	25	9.5	Good	Opinions
Sea Otter	21	24	7.5	Poor	RN flight tests
PH-3	22.5	22.5	7.7	Good	Opinion
PBY-5	20	20	8.2	Fair	Opinions and RAF flight tests
Singapore II	26	30.5	8.6	Good	RAF flight tests
PBN-1	22.5	22.5	8.4	Fair	NAF flight tests
Singapore III	26	30.5	8.6	Good	RAF flight tests
Lerwiok Ia	30	33.5	7.3	Poor	RAF flight tests
Lerwiok Ib	30	33.5	8.0	Good	RAF flight tests
Sunderland I	31	41	9.4	Good	RAF flight tests
31	20	20	6.7	Poor	Consolidated flight tests
XP4Y-1	20	20	8.0	Fair	Consolidated flight tests
PBM-3	20	30	8.5	Good	Martin and NACA flight tests and opinions
Sunderland III	31	41	8.8	Good	RAF flight tests
PBM-5	20	30	8.5	Fair	Opinions
XPBB-1	20	20	7.6	Good	Navy flight tests and NACA RB No. 3I27
PB2Y-3	22.5	22.5	7.5	Poor	Consolidated, Navy, and RAF flight tests
G	31	41	9.5	Fair	RAF flight tests
314	22.5	22.5	6.9	Poor	NACA RB No. 3I27
A-314	22.5	22.5	7.6	Fair	NACA RB No. 3I27
XPB2M-1R	20	20	8.0	Fair	Navy flight tests and opinions

¹ Measured at step on forebody and at maximum dead rise on afterbody



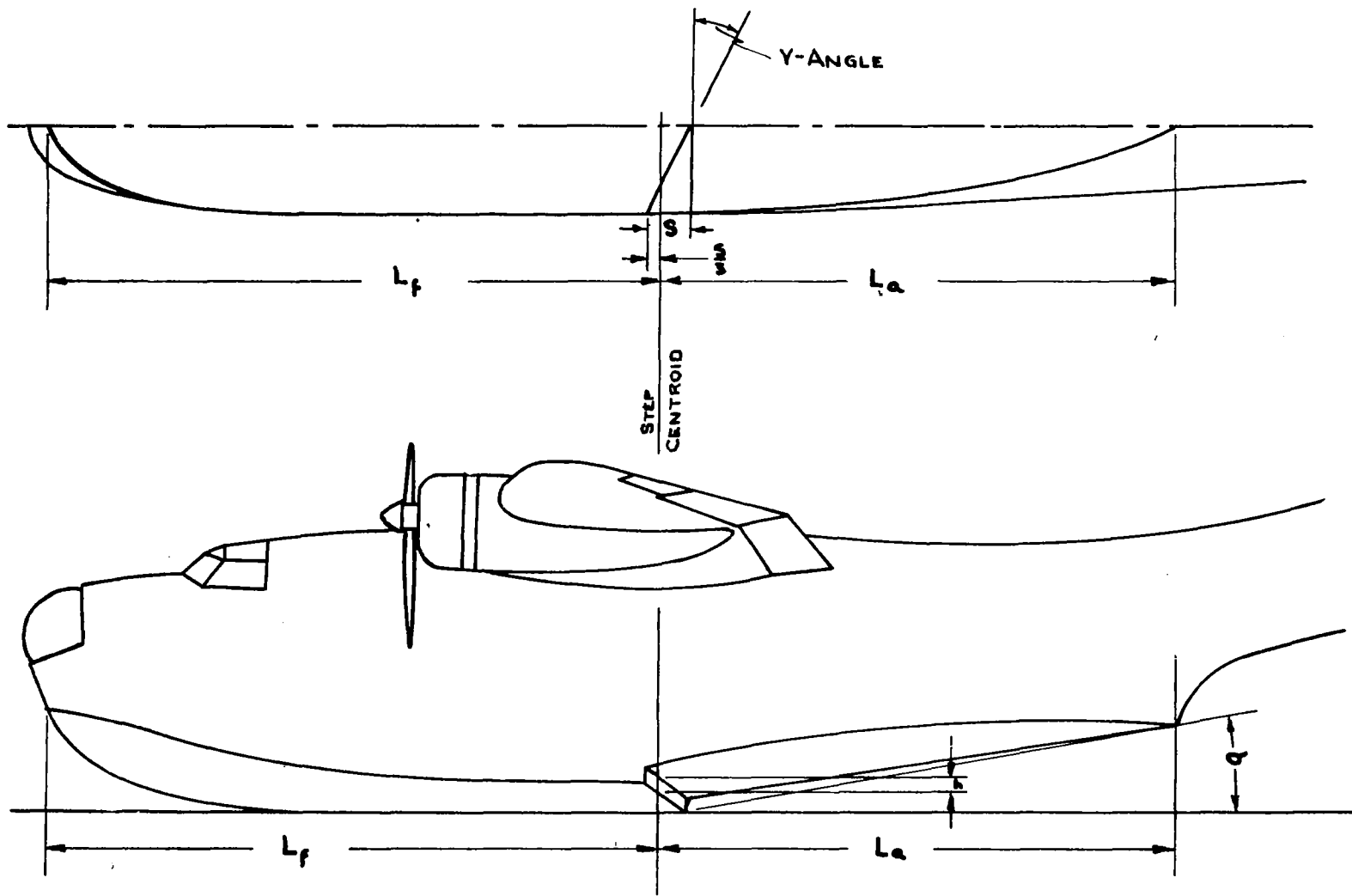
V-STEP USED ON THE SHORT "EMPIRE"

Note the uniform depth of the face of the step obtained by using greater deadrise on the afterbody than on the forebody.



V-STEP USED ON THE PEN-1

Note the throttle in the step face resulting from using the same deadrise on the afterbody and forebody.



SCHEMATIC SKETCH TO ILLUSTRATE
DEFINITIONS OF DIMENSIONS

FIG. 2

FIG. 2

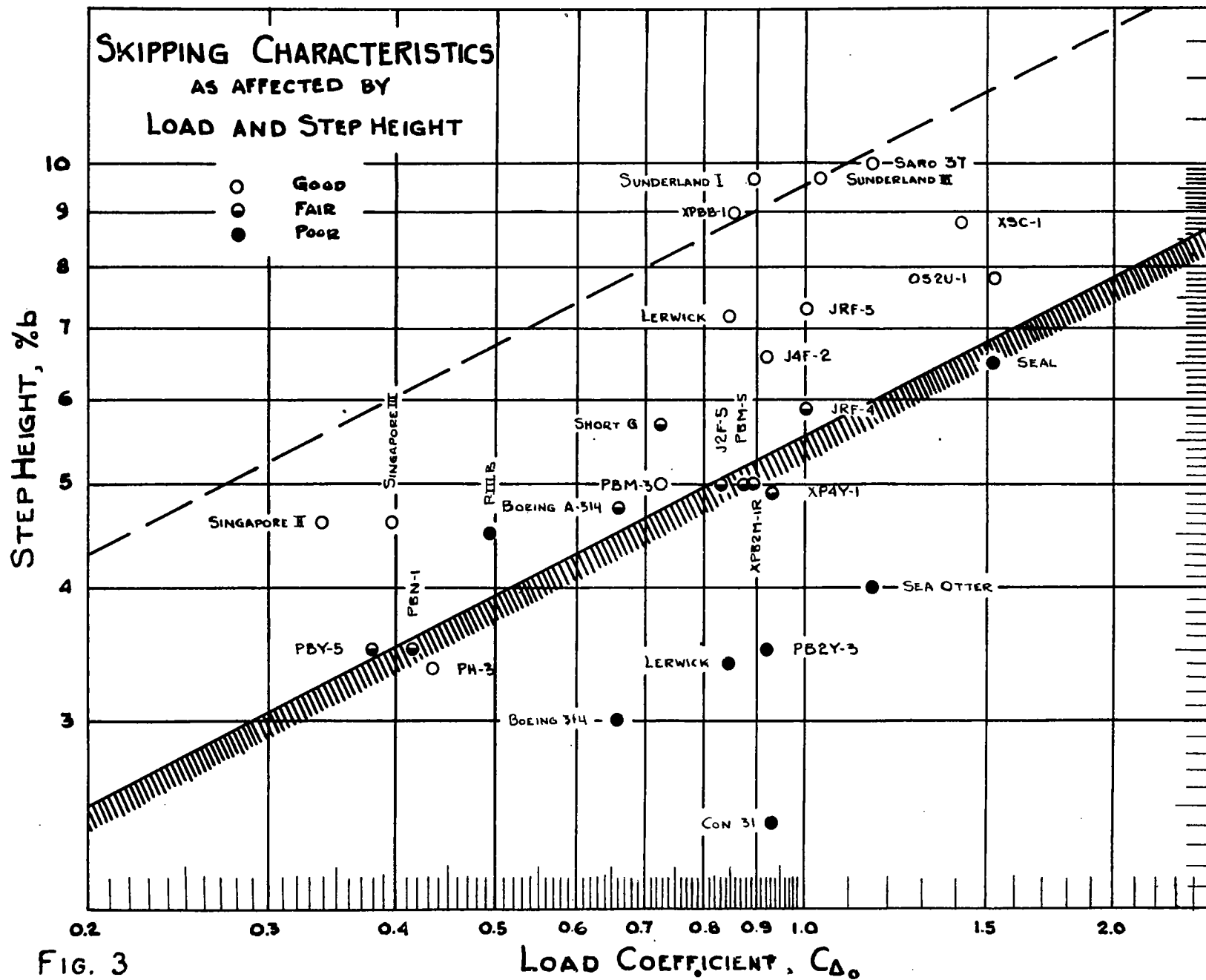
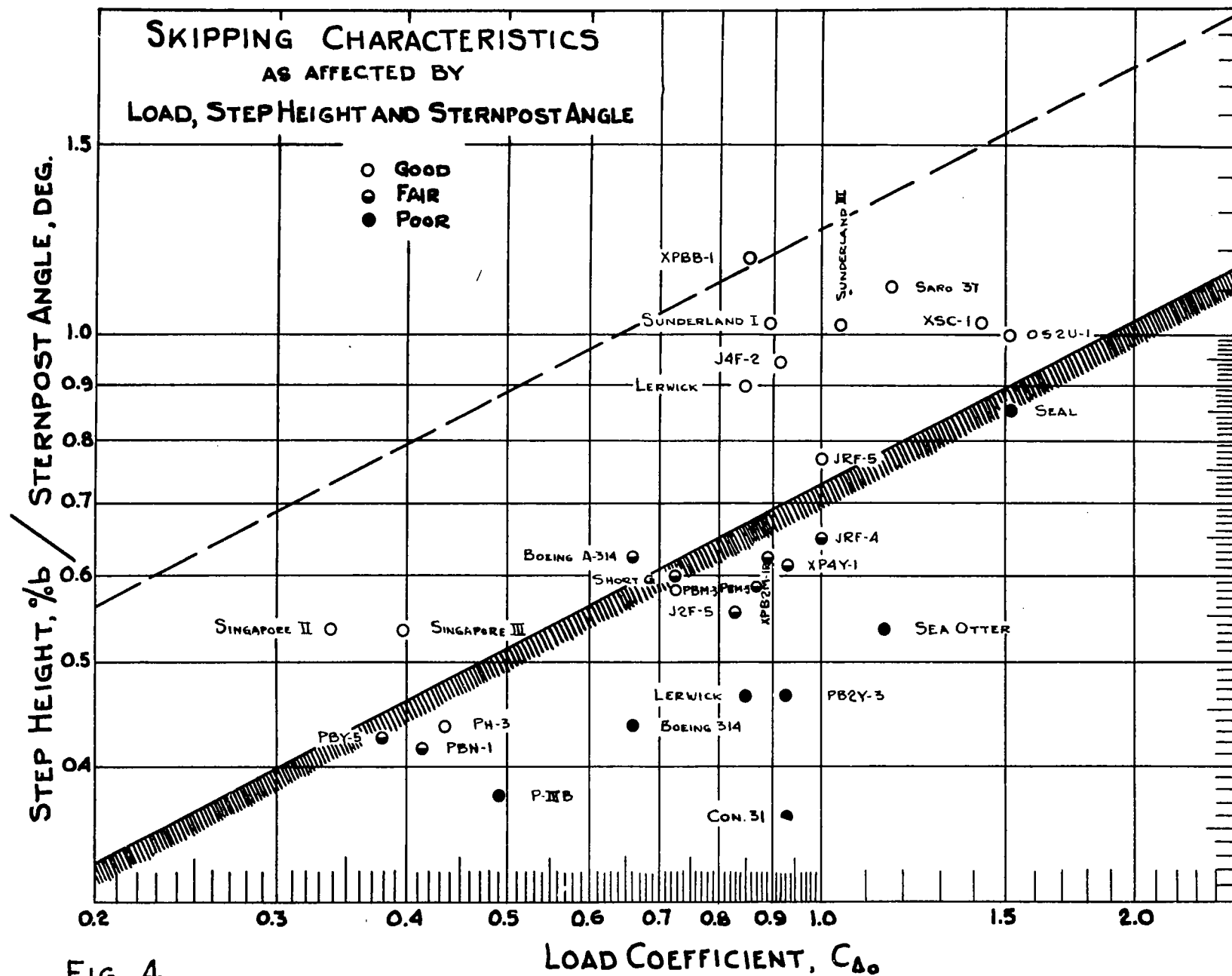


FIG. 3

FIG. 3



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